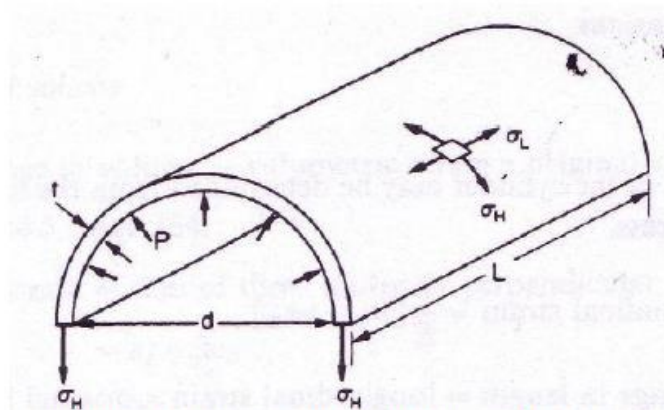


Question 1



a.

Given the above structure which shows a freebody diagram of part of a thin cylinder subjected to internal pressure show that the hoop stress (σ_H) is given by the following relationship.

$$\sigma_H = \frac{Pd}{2t}$$

[4 marks]

- b. A plane fuselage which is circular in cross section, is made from Alu7075 and the hoop stress must not exceed 200MPa. The fuselage is 3.5m in diameter and has a skin thickness of 0.5mm. Give that $P = \rho gh$ using table 1 determine to the nearest 5000ft the maximum altitude that this aircraft can fly at.

- i. If the cabin is sealed at sea level

[4 marks]

- ii. If the cabin is pressurized to an equivalent pressure of 8000ft

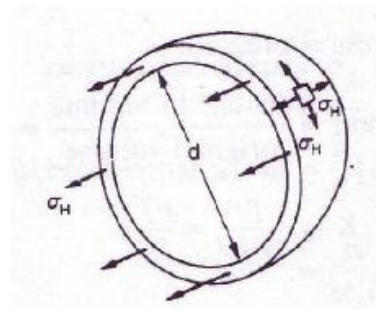
[3 Marks]

- c. In terms of the plane structure, using your answer to part b comment on the advantage of pressurizing plane cabins.

[2 Marks]

- d. Given the below structure which shows a freebody diagram of part of a sphere subjected to internal pressure show that the hoop stress (σ_H) is given by the following relationship.

$$\therefore \sigma_H = \frac{pd}{4t}$$



- e. Based on the equations derived in parts a and d, discuss why in oil storage facilities some fluids are stored in spherical tanks whilst others are stored in cylindrical containers.

[3 Marks]

Table 1

Altitude Above Sea Level		Absolute Barometer		Absolute Atmospheric Pressure		
feet	meters	inches Hg	mm Hg	psia	kg/cm ²	kPa
-5000	-1524	35.7	908	17.5	1.23	121
-4500	-1372	35.1	892	17.2	1.21	119
-4000	-1219	34.5	876	16.9	1.19	117
-3500	-1067	33.9	861	16.6	1.17	115
-3000	-914	33.3	846	16.4	1.15	113
-2500	-762	32.7	831	16.1	1.13	111
-2000	-610	32.1	816	15.8	1.11	109
-1500	-457	31.6	802	15.5	1.09	107
-1000	-305	31.0	788	15.2	1.07	105
-500	-152	30.5	774	15.0	1.05	103
0 ¹⁾	0	29.9	760	14.7	1.03	101
500	152	29.4	746	14.4	1.01	99.5
1000	305	28.9	733	14.2	0.997	97.7
1500	457	28.3	720	13.9	0.979	96.0
2000	610	27.8	707	13.7	0.961	94.2
2500	762	27.3	694	13.4	0.943	92.5
3000	914	26.8	681	13.2	0.926	90.8
3500	1067	26.3	669	12.9	0.909	89.1
4000	1219	25.8	656	12.7	0.893	87.5
4500	1372	25.4	644	12.5	0.876	85.9
5000	1524	24.9	632	12.2	0.860	84.3
6000	1829	24.0	609	11.8	0.828	81.2
7000	2134	23.1	586	11.3	0.797	78.2
8000	2438	22.2	564	10.9	0.768	75.3
9000	2743	21.4	543	10.5	0.739	72.4
10000	3048	20.6	523	10.1	0.711	69.7
15000	4572	16.9	429	8.29	0.583	57.2
20000	6096	13.8	349	6.75	0.475	46.6
25000	7620	11.1	282	5.45	0.384	37.6
30000	9144	8.89	226	4.36	0.307	30.1
35000	10668	7.04	179	3.46	0.243	23.8
40000	12192	5.52	140	2.71	0.191	18.7
45000	13716	4.28	109	2.10	0.148	14.5
50000	15240	3.27	83	1.61	0.113	11.1